Spacecraft Separation Analysis Tool



 Team Members: Michael Clopton, Viviana Gallardo-Cardiel, Alejandro Hernandez, Julianna Hernandez, Kishan Patel Faculty Advisor: Ray Manning
The Aerospace Corporation Liaison: Edgar Herrera Department of Mechanical Engineering
College of Engineering, Computer Science, and Technology California State University, Los Angeles



Project Background

The most critical function of a launch vehicle is placing a payload into its proper mission orbit. The Aerospace Corporation is interested in developing a rapid turnaround software tool for the analysis, optimization, and uncertainty analysis of the separation system and its effects on the resulting payload trajectory.

Project Objective

• Provide a software tool that performs rapid and parametric analysis of a spacecraft separation system.

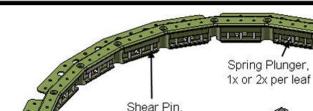
Mars 2020

Spacecraft

Requirement		Performance Objectives
Software Framework	Customer Software Cost	Minimum cost suggests open source framework
	User Interface	Does not need to run over internet
	Simulation Architecture	Single upper stage and single payload
	Optimization	Separation system spring radius, angle, constant, undeformed length
	Uncertainty Analysis	Payload & upper stage mass, inertias, center of mass offsets
Separation Rate	Min Axial Separation	0.6 m/s
	Max Pitch Rate	0.01 rad/s
	Max Yaw Rate	0.01 rad/s
Documentation	User Manual	Details instructions on how to use simulation analysis tool

Project Requirements

Overall Design Approach



Read Input Parameters

Read

• Payload, Upper stage, Separation system, Optimization & Time history

